

**DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION**

A00066CE Revision 0 Costruzioni Aeronautiche Tecnam srl P2010 December 4, 2015
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TYPE CERTIFICATE DATA SHEET No. A00066CE

This Data Sheet, which is part of Type Certificate No. A00066CE, prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder Costruzioni Aeronautiche Tecnam srl
 Via Tasso, 478
 80127 Napoli
 Italy

I - Model P2010 (Normal Category), Approved December 4, 2015

Engine Lycoming Engine IO-360-M1A (TC 1E10)

Fuel AVGAS Grade 91/96 or 100LL (ASTM D910)

Engine Limits Max rotational speed 2700 r.p.m. (180 hp)
 Cruise rotational speed (75%) 2450 r.p.m. (135 hp)
 (Engine shaft r.p.m)

Propeller and Propeller Limits MT Propeller MT 188R145-4G (TC P19BO)
 Two blades, constant speed, fixed pitch, wood construction. Diameter: 1880 mm (74 in) - no reduction allowed.
 Clockwise rotation (pilot's view)

Oil

Average Ambient Temperature	MIL-L-6082B or SAEJ1966 Spec. Mineral Grades	MIL-L-22851 or SAEJ1899 Spec. Ashless Dispersant Grades
All Temperatures		SAE15W50 or SAE20W-50
Above 80°F	SAE60	SAE60
Above 60°F	SAE50	SAE40 or SAE50
30°F to 90°F	SAE40	SAE40
0°F to 70°F	SAE30	SAE40, SAE30, SAE20W40
Below 10°F	SAE20	SAE30 or SAE20W30

For additional info, refer to "Lycoming Operation and Installation Manual" for the list of alternative recommended commercial brands and types

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		KIAS	KCAS
Airspeed Limits	VO (Operating Manoeuvring Speed)	120	119
	VA (Design Manoeuvring Speed)	120	119
	VFE (Maximum Flap Extended Speed)	91	92
	VNO (Maximum Structural Cruising Speed)	132	130
	VNE (Never Exceed Speed)	166	164
Center of Gravity (C.G.) Range	Forward limit:10.31 in (19 % MAC) behind datum Aft Limit: 17.32 in (32.0 % MAC) behind datum Mean Aerodynamic Chord is 54.2 in		
Empty Weight C.G. Range	None		
Datum:	Vertical plane tangent to wing leading edge		
Levelling Means	Seat track supporting beams (see AFM, 2010/100, Sect.6 for the procedure).		
Maximum Weight	Take-off	2557 lbs	
	Landing	2557 lbs	
Minimum Crew	1 pilot		
Number of Seats	4		
Maximum Compartments Weights	88 lbs at 61.41 in aft the datum		
Fuel Capacity	63.4 US Gal (+ 24.1 in) usable 61 US Gal		
Oil Capacity (each engine)	Maximum:	8 US Qts	
	Minimum:	4 US Qts	
Control Surface Movements (*)	Ailerons	19°±2° TEU(**)	14 ° ±2° TED(***)
	Stabilator	6°±2° TEU	17°±2° TED
	Stabilator trim tab	3°±1° TEU	15°±1° TED
	Rudder	25°±2° RH	25°±2° LH
	Rudder trim tab	20°±2° RH	20°±2° LH
	Flaps	15°±1° TED (Take-off position)	
		40°±1° TED (Landing position)	
	(*) Nominal Values (**) Trailing Edge Up (***) Trailing Edge Down		
Applicable Serial Numbers	S/N I/US to 9999/US		
Import Requirements	a) A U.S. airworthiness certificate may be issued on the basis of an NAA Export Certificate of Airworthiness (Export of C of A) signed by a representative of the Ente Nazionale per l'Aviazione Civile (ENAC) on behalf of the European Community. The Export C of A should contain the following statement "The aircraft covered by this certificate has been examined, tested, and found to comply with U.S. Type Certificate No. A00066CE and to be in a condition for safe operation."		
	b) The U.S. airworthiness certification basis for aircraft type certificated under 14 CFR part 21, section 21.29 and exported		

- c) Each P2010 aircraft should have the following modification installed:
- MOD2010/061 “New cargo net and G1000 software configuration for USA aircraft”,

It must be identified with a “Steel identification plate” showing USA S/N (xxx/US) and TCDS references. Tecnam can incorporate these modifications using Tecnam Service Bulletin SB 221-CS.

Certification Basis

Type Certification under 14 CFR Section 21.29 including the following requirements:
 14 CFR Part 23 effective February 1, 1965 including amdt 23-1 through 23-61
 14 CFR Part 36 effective December 1, 1969 including amdt 36-1 through 36-28

Equivalent levels of safety (ELOS): None

Approved Kinds of Operation

Day and Night, Visual Flight Rules (VFR) and Instrument Flight Rules (IFR)

Prohibited Kinds of Operation

Flight into known icing conditions

Type Certificate No. A00066CE was issued December 4, 2015.
 Date of Application for FAA Type Certificate was December 13, 2011.

The European Aviation Safety Agency (EASA) originally type certified this aircraft under its type certificate number A.576.

Maximum Operating Altitude

12000 ft

Equipment

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the airplane for certification. Such equipment is listed in the current FAA approved Airplane Flight Manual: 2010/100 ed2 rev0 or later approved revisions.

Service Information

Each of the documents listed below must state that it is approved by the European Aviation Safety Agency (EASA):

- Service bulletins,
- Structural repair manuals,
- Vendor manuals,
- Aircraft flight manuals, and
- Overhaul and maintenance manuals.

The FAA accepts such documents and considers them FAA-approved for type design data only unless one of the following conditions exists:

- The documents change the limitations, performance, or procedures of the FAA approved manuals; or
- The documents make an acoustical or emissions changes to this product’s U.S. type certificate as defined in 14 CFR § 21.93.

The FAA uses the post type validation procedures to approve these documents. The FAA may delegate on case-by-case to EASA to approve on behalf of the FAA for the U.S. type certificate. If this is the case it will be noted on the document.

Each airplane is provided with the following approved documents:

- a) AFM doc. 2010/100 ed2 rev.0, or later FAA approved revision.
- b) Airplane Maintenance Manual doc. 2010/101 ed1 rev2, or later FAA approved revision, including Chap. 4: "Airworthiness Limitations" and Chap. 5: "Time Limits/ Maintenance Check".
- c) The appropriate Lycoming LIO-360-M1A series engine maintenance manuals.
- d) The appropriate Instruction Manual MT Propellers Doc No. E-112

NOTES:

NOTE 1

Current weight and balance report, including list of equipment included in certificated empty weight and loading instructions when necessary, must be provided for each aircraft at the time of original certification.

The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel 14.28 lb. at 24.1 in. aft of datum

NOTE 2

Airplane operation must be in accordance with the EASA approved Airplane Flight Manual listed above. All placards listed in Section 2 must be displayed.

NOTE 3

Airworthiness Limitations are specified in the Section 2 LIMITATIONS of the Flight Manual and Chapter 4 of the Instructions for Continued Airworthiness (Maintenance Manual) and are approved by the EASA and the FAA. These LIMITATIONS specify mandatory replacement times, and operating limitations, and may not be changed without FAA approval.

Revisions to the Airworthiness Limitations must be approved by the FAA. The inspections, maintenance, repair and painting must be accomplished according to the Maintenance Manual or other procedures acceptable to the FAA.

NOTE 4

Information essential for the proper operation, maintenance and inspection of the airplane is contained in the Tecnam P2010 Airplane Flight Manual and Maintenance Manual.

NOTE 5

Optional Variable Pitch Propeller

Tecnam Modification No. MOD2010-002 (Variable Pitch Propeller - MTV-15-B/193-52 (TC P23BO)). Airplanes with this modification must have at least Tecnam AFM doc. 2010/100 ed2 rev.0 Supplement No. D-2 or later FAA/EASA approved revisions and Tecnam P2010 Aircraft Maintenance Manual Supplement No. S-1, 1 Edition, Rev. 0, or later FAA/EASA approved revisions.

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